

§ 23.171

(c) *Longitudinal trim.* The airplane must maintain longitudinal trim under each of the following conditions:

- (1) A climb with—
 - (i) Takeoff power, landing gear retracted, wing flaps in the takeoff position(s), at the speeds used in determining the climb performance required by § 23.65; and
 - (ii) Maximum continuous power at the speeds and in the configuration used in determining the climb performance required by § 23.69(a).
- (2) Level flight at all speeds from the lesser of V_H and either V_{NO} or V_{MO}/M_{MO} (as appropriate), to $1.4 V_{S1}$, with the landing gear and flaps retracted.
- (3) A descent at V_{NO} or V_{MO}/M_{MO} , whichever is applicable, with power off and with the landing gear and flaps retracted.
- (4) Approach with landing gear extended and with—
 - (i) A 3 degree angle of descent, with flaps retracted and at a speed of $1.4 V_{S1}$;
 - (ii) A 3 degree angle of descent, flaps in the landing position(s) at V_{REF} ; and
 - (iii) An approach gradient equal to the steepest used in the landing distance demonstrations of § 23.75, flaps in the landing position(s) at V_{REF} .
- (d) In addition, each multiple airplane must maintain longitudinal and directional trim, and the lateral control force must not exceed 5 pounds at the speed used in complying with § 23.67(a), (b)(2), or (c)(3), as appropriate, with—
 - (1) The critical engine inoperative, and if applicable, its propeller in the minimum drag position;
 - (2) The remaining engines at maximum continuous power;
 - (3) The landing gear retracted;
 - (4) Wing flaps retracted; and
 - (5) An angle of bank of not more than five degrees.
- (e) In addition, each commuter category airplane for which, in the determination of the takeoff path in accordance with § 23.57, the climb in the takeoff configuration at V_2 extends beyond 400 feet above the takeoff surface, it must be possible to reduce the longitudinal and lateral control forces to 10 pounds and 5 pounds, respectively, and the directional control force must not exceed 50 pounds at V_2 with—

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- (1) The critical engine inoperative and its propeller in the minimum drag position;
- (2) The remaining engine(s) at takeoff power;
- (3) Landing gear retracted;
- (4) Wing flaps in the takeoff position(s); and
- (5) An angle of bank not exceeding 5 degrees.

[Doc. No. 4080, 29 FR 17955, Dec. 18, 1964, as amended by Amdt. 23–21, 43 FR 2318, Jan. 16, 1978; Amdt. 23–34, 52 FR 1828, Jan. 15, 1987; Amdt. 23–42, 56 FR 351, Jan. 3, 1991; 56 FR 5455, Feb. 11, 1991; Amdt. 23–50, 61 FR 5189, Feb. 9, 1996]

STABILITY

§ 23.171 General.

The airplane must be longitudinally, directionally, and laterally stable under §§ 23.173 through 23.181. In addition, the airplane must show suitable stability and control “feel” (static stability) in any condition normally encountered in service, if flight tests show it is necessary for safe operation.

§ 23.173 Static longitudinal stability.

Under the conditions specified in § 23.175 and with the airplane trimmed as indicated, the characteristics of the elevator control forces and the friction within the control system must be as follows:

- (a) A pull must be required to obtain and maintain speeds below the specified trim speed and a push required to obtain and maintain speeds above the specified trim speed. This must be shown at any speed that can be obtained, except that speeds requiring a control force in excess of 40 pounds or speeds above the maximum allowable speed or below the minimum speed for steady unstalled flight, need not be considered.
- (b) The airspeed must return to within the tolerances specified for applicable categories of airplanes when the control force is slowly released at any speed within the speed range specified in paragraph (a) of this section. The applicable tolerances are—
 - (1) The airspeed must return to within plus or minus 10 percent of the original trim airspeed; and

(2) For commuter category airplanes, the airspeed must return to within plus or minus 7.5 percent of the original trim airspeed for the cruising condition specified in § 23.175(b).

(c) The stick force must vary with speed so that any substantial speed change results in a stick force clearly perceptible to the pilot.

[Doc. No. 4080, 29 FR 17955, Dec. 18, 1964, as amended by Amdt. 23-14, 38 FR 31820 Nov. 19, 1973; Amdt. 23-34, 52 FR 1828, Jan. 15, 1987]

§ 23.175 Demonstration of static longitudinal stability.

Static longitudinal stability must be shown as follows:

(a) *Climb*. The stick force curve must have a stable slope at speeds between 85 and 115 percent of the trim speed, with—

- (1) Flaps retracted;
- (2) Landing gear retracted;
- (3) Maximum continuous power; and
- (4) The airplane trimmed at the speed used in determining the climb performance required by § 23.69(a).

(b) *Cruise*. With flaps and landing gear retracted and the airplane in trim with power for level flight at representative cruising speeds at high and low altitudes, including speeds up to V_{NO} or V_{MO}/M_{MO} , as appropriate, except that the speed need not exceed V_H —

(1) For normal, utility, and acrobatic category airplanes, the stick force curve must have a stable slope at all speeds within a range that is the greater of 15 percent of the trim speed plus the resulting free return speed range, or 40 knots plus the resulting free return speed range, above and below the trim speed, except that the slope need not be stable—

- (i) At speeds less than $1.3 V_{S1}$; or
- (ii) For airplanes with V_{NE} established under § 23.1505(a), at speeds greater than V_{NE} ; or
- (iii) For airplanes with V_{MO}/M_{MO} established under § 23.1505(c), at speeds greater than V_{FC}/M_{FC} .

(2) For commuter category airplanes, the stick force curve must have a stable slope at all speeds within a range of 50 knots plus the resulting free return speed range, above and below the trim speed, except that the slope need not be stable—

- (i) At speeds less than $1.4 V_{S1}$; or

(ii) At speeds greater than V_{FC}/M_{FC} ; or

(iii) At speeds that require a stick force greater than 50 pounds.

(c) *Landing*. The stick force curve must have a stable slope at speeds between $1.1 V_{S1}$ and $1.8 V_{S1}$ with—

- (1) Flaps in the landing position;
- (2) Landing gear extended; and
- (3) The airplane trimmed at—

(i) V_{REF} , or the minimum trim speed if higher, with power off; and

(ii) V_{REF} with enough power to maintain a 3 degree angle of descent.

[Doc. No. 27807, 61 FR 5190, Feb. 9, 1996]

§ 23.177 Static directional and lateral stability.

(a)(1) The static directional stability, as shown by the tendency to recover from a wings level sideslip with the rudder free, must be positive for any landing gear and flap position appropriate to the takeoff, climb, cruise, approach, and landing configurations. This must be shown with symmetrical power up to maximum continuous power, and at speeds from $1.2 V_{S1}$ up to V_{FE} , V_{LE} , V_{NO} , V_{FC}/M_{FC} , whichever is appropriate.

(2) The angle of sideslip for these tests must be appropriate to the type of airplane. The rudder pedal force must not reverse at larger angles of sideslip, up to that at which full rudder is used or a control force limit in § 23.143 is reached, whichever occurs first, and at speeds from $1.2 V_{S1}$ to V_O .

(b)(1) The static lateral stability, as shown by the tendency to raise the low wing in a sideslip with the aileron controls free, may not be negative for any landing gear and flap position appropriate to the takeoff, climb, cruise, approach, and landing configurations. This must be shown with symmetrical power from idle up to 75 percent of maximum continuous power at speeds from $1.2 V_{S1}$ in the takeoff configuration(s) and at speeds from $1.3 V_{S1}$ in other configurations, up to the maximum allowable airspeed for the configuration being investigated (V_{FE} , V_{LE} , V_{NO} , V_{FC}/M_{FC} , whichever is appropriate) in the takeoff, climb, cruise, descent, and approach configurations. For the landing configuration, the power must